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	Application No.	Applicant(s)	
Notice of Allowability	10/615,094	ACKERMAN ET AL	
	Examiner	Art Unit	
	Timothy H. Meeks	1762	
The MAILING DATE of this communication appearable communication appearable claims being allowable, PROSECUTION ON THE MERITS IS (herewith (or previously mailed), a Notice of Allowance (PTOL-85) of NOTICE OF ALLOWABILITY IS NOT A GRANT OF PATENT RICE of the Office or upon petition by the applicant. See 37 CFR 1.313	(OR REMAINS) CLOSED in this apport of the communication (OHTS). This application is subject to and MPEP 1308.	pplication. If not include n will be mailed in due	ed course. THIS
1. This communication is responsive to the amendment filed o	<u>)n 12/30/04</u> .		
2. ☑ The allowed claim(s) is/are <u>1-11 and 13-30</u> .			
3. \boxtimes The drawings filed on <u>08 July 2003</u> are accepted by the Exa	aminer.		
4.			
Attachment(s) 1. ☐ Notice of References Cited (PTO-892) 2. ☐ Notice of Draftperson's Patent Drawing Review (PTO-948) 3. ☐ Information Disclosure Statements (PTO-1449 or PTO/SB/08 Paper No./Mail Date 4. ☐ Examiner's Comment Regarding Requirement for Deposit of Biological Material	5. ☐ Notice of Informal P 6. ☐ Interview Summary Paper No./Mail Dat 7. ☑ Examiner's Amendr 8. ☑ Examiner's Stateme 9. ☑ Other	(PTO-413), te ment/Comment	·

EXAMINER'S AMENDMENT

An examiner's amendment to the record appears below. Should the changes and/or additions be unacceptable to applicant, an amendment may be filed as provided by 37 CFR 1.312. To ensure consideration of such an amendment, it MUST be submitted no later than the payment of the issue fee.

Authorization for this examiner's amendment was given in a telephone interview with Donald Hasse on 2/15/05.

The application has been amended as follows:

Claim 1 has been amended as follows:

- 1. (Currently Amended.) A method for forming a controlled, relatively uniform aluminide coating on a turbine engine component having an external surface and an internal cavity defined by an internal surface that is connected to the external surface by at least one hole, the method being conducted in a vapor coating container having a hollow interior coating chamber, the method comprising the steps of:
 - (a) loading the coating chamber with the component to be coated;
- (b) heating the loaded coating chamber to a temperature of from about 240°C to about 450°C;
- (c) flowing a tri-alkyl aluminum coating gas into the heated coating chamber at a pressure of from about 50 to about 2000 mtorr (about 0.68 to about 27 kgf/m²) for from about 0.25 to about 4 hours to deposit an aluminum coating on the external and internal surfaces of the component; and then

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(d) heating the coated component in a a nonoxidizing atmosphere to a

temperature of from about 500°C to about 1100°C to form an aluminide coating on the

external and internal surfaces of the component,

wherein the aluminide coating on the internal surface of the blade is less than

about 0.003 inches (less than about 76.2 microns) thick in the airfoil body portion and

less than about 0.0015 inches (less than about 38.1 microns) thick in the root portion.

Claim 12 has been cancelled.

In claim 13, line 1, "12" has been changed to --1--.

Claim 17 has been amended as follows:

17. (Currently Amended.) A method for forming a controlled, relatively uniform

aluminide coating on a turbine engine blade having an external surface and an internal

cooling cavity defined by an internal surface that is connected to the external surface by

cooling holes, the method being conducted in a vapor coating container having a hollow

interior coating chamber, the method comprising the steps of:

(a) loading the coating chamber with the blade to be coated;

(b) heating the loaded coating chamber to a temperature of from about 240°C to

about 450°C;

(c) flowing a tri-alkyl aluminum coating gas into the heated coating chamber at a pressure of from about 50 to about 2000 mtorr (about 0.68 to about 27 kgf/m²) for from about 0.25 to about 4 hours to deposit an aluminum coating on the external and internal surfaces of the blade;

- (d) heating the coated blade in a nonoxidizing atmosphere to a temperature of from about 500°C to about 1100°C to form an aluminide coating on the external and internal surfaces of the blade; and then
- (e) maintaining the blade at a temperature of from about 450°C to about 1100°C in the presence of oxygen to form an oxide coating on the external and internal surfaces of the blade;

wherein the aluminide coating [has a thickness of from about 0.0015 to about 0.003 inches (from about 38.1 to about 76.2 microns) on the external surface of the blade in the airfoil body portion, and has a thickness of from about 0.0005 to about 0.0015 inches (from about 12.7 to about 38.1 microns) on the internal surface of the blade] on the internal surface of the blade is less than about 0.003 inches (less than about 76.2 microns) thick in the airfoil body portion and less than about 0.0015 inches (less than about 38.1 microns) thick in the root portion.

Claim 28 has been amended as follows:

28. (Currently Amended) A method for forming a controlled, relatively uniform aluminide coating on a turbine engine blade having an external surface and an internal cooling cavity defined by an internal surface that is connected to the external

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surface by cooling holes, the method being conducted in a vapor coating container having a hollow interior coating the chamber. the method comprising the steps of:

- (a) loading the coating chamber with the blade to be coated;
- (b) heating the loaded coating chamber to a temperature of from about 250°C to about 300°C;
- (c) flowing a tri-alkyl aluminum coating gas into the heated coating chamber at a pressure of from about 450 to about 550 mtorr (about 6.1 to about 7.5 kgf/m²) for from about 0.5 to about 2 hours to deposit an aluminum coating on the external and internal surfaces of the blade;
- (d) heating the coated blade in a vacuum to a temperature of from about 640°C to about 700°C to form aluminide coating on the external and internal surfaces of the blade; and then
- (e) maintaining the blade at a temperature of from about 600°C to about 800°C in the presence of oxygen to form an oxide coating on the external and internal surfaces of the blade

wherein the aluminide coating on the internal surface of the blade is less than about 0.003 inches (less than about 76.2 microns) thick in the airfoil body portion and less than about 0.0015 inches (less than about 38.1 microns) thick in the root portion.

The following is an examiner's statement of reasons for allowance: The prior art does not disclose or fairly suggest providing the different claimed ranges of thicknesses

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of the aluminide coating on the internal surfaces of the body versus the internal surfaces of the root portion of the blade. As set forth by applicants at paragraph 0026 of the specification, limiting the thickness of the coating in the root portion to the claimed range prevents material degradation and cracking in the root portion.

Any comments considered necessary by applicant must be submitted no later than the payment of the issue fee and, to avoid processing delays, should preferably accompany the issue fee. Such submissions should be clearly labeled "Comments on Statement of Reasons for Allowance."

Any inquiry concerning this communication or earlier communications from the examiner should be directed to Timothy H. Meeks whose telephone number is (571) 272-1423. The examiner can normally be reached on Mon 6-6 and Tues-Thurs 9-5.

If attempts to reach the examiner by telephone are unsuccessful, the examiner's supervisor, Shrive P. Beck can be reached on (571) 272-1415. The fax phone number for the organization where this application or proceeding is assigned is 703-872-9306.

Information regarding the status of an application may be obtained from the Patent Application Information Retrieval (PAIR) system. Status information for published applications may be obtained from either Private PAIR or Public PAIR. Status information for unpublished applications is available through Private PAIR only. For more information about the PAIR system, see http://pair-direct/uspto.gov. Should you have questions on access to the Private PAIR system, contact the Electronic Business Center (EBC) at 866-217-9197 (toll-free).

Timothy H Meeks Primary Examiner Art Unit 1762